

# Recommended Practice for Pressure Measurements and Calculation of Effective Pumping Speeds during Electric Propulsion Testing

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**The electric propulsion community has been implored to establish and implement a set of universally applicable test standards during the research, development, and qualification of electric propulsion systems. Variability between facility-to-facility and more importantly ground-to-flight performance can result in large margins in application or aversion to mission infusion. Performance measurements and life testing under appropriate conditions can be costly and lengthy. Measurement practices must be consistent, accurate, and repeatable. Additionally, the measurements must be universally transportable across facilities throughout the development, qualification, spacecraft integration, and on-orbit performance. A recommended practice for making pressure measurements, pressure diagnostics, and calculating effective pumping speeds with justification is presented.**

## Nomenclature

$A_e$	= cross-sectional area of the electron beam
$A_\sigma$	= collective ionization cross area
$e$	= electron charge
$I_C$	= collection current
$k$	= pumping speed reduction factor
$k$	= Boltzmann constant = $1.380648 \times 10^{-23}$ J/K
$L$	= Length
$N_v$	= number of moles in the volume
$n$	= molecular density
$\sigma_i$	= ionization cross section
$p$	= pressure

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$R_g$	=	gauge relative sensitivity factor
$S$	=	gauge sensitivity factor
$T$	=	temperature
$V_g$	=	grid potential

## I. Introduction

The acknowledgement of challenges associated with accurate, consistent, repeatable, and transportable test results of electric propulsion systems has been well documented in recent years.<sup>1</sup> As such, there has been significant interest in developing rigorous standards for electric propulsion testing from both internal and external communities. Several community efforts including the European Space Agency (ESA)<sup>2</sup>, the Joint Army Navy NASA Air Force (JANNAF) Committee on Electric Propulsion Diagnostics, and the American Institute of Aeronautics and Astronautics (AIAA) have begun efforts to “standardize” practices and measurements during the testing of electric propulsion devices.

Variability between facility-to-facility and more importantly ground-to-flight performance can result in large margins in application or aversion to mission infusion. Performance measurements and life testing under appropriate conditions can be costly and lengthy. Measurement practices must be consistent, accurate, and repeatable. Additionally, the measurements must be universally transportable across facilities throughout the development, qualification, spacecraft integration and on-orbit performance. Critical to transportable measurements is the understanding and influence of facility effects. Background pressure is one of the primary facility effects well documented to influence ground test results.<sup>3</sup>

This manuscript describes a recommended practice for measuring pressure and calculating neutral density and effective pumping speeds during electric propulsion testing. A universal theory for an a priori prediction of the influence of facility effects for all thrusters is incomplete. Consequently, these recommendations are intended to evolve as the community continues to develop new theories, modeling capabilities and practices.

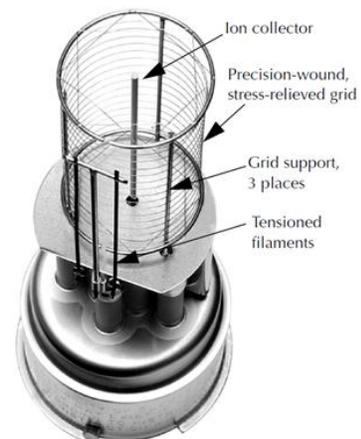
## II. Pressure Gauge Description and Operation

This section summarizes a recommended gauge design and practices for the purpose of pressure measurements and neutral density calculations in vacuum systems over the range of interest of electric propulsion testing. This is acknowledged only to be a first iteration with the community, and applicability is intentionally limited to gridded-ion and Hall thrusters. The pressure measurements of the recommended gauge are applicable over the general range of  $10^{-8}$  to  $10^{-4}$  Torr for  $N_2$  or air.

Vacuum pressure gauges measure pressure from  $10^{-16}$  Torr to atmospheric. No single measurement technique can accurately measure over this entire range. Instead, various types of gauges are available for the specific range of interest. Higher pressure ranges can be accommodated with techniques relying on mechanical deflection or thermal conductivities including thermocouple gauges, diaphragms, liquid manometers, etc. In the lower pressure range, measurement techniques often rely on the collection of ions generated from the ambient gas to measure the neutral density, which is then converted to pressure. High-vacuum gauges include cold cathode ionization gauges, hot cathode (Bayard-Alpert) ionization gauges, and spinning rotor gauges. Description and comparison of the various gauges is available in open literature. For brevity, this document is limited to the recommend gauge; the Bayard-Alpert (BA) hot cathode ionization gauge.

### A. Bayard-Alpert Hot Cathode Gauge Theory

All BA gauges consist of a heated cathode filament, an acceleration grid, and a grounded collector; as shown in Fig. 1. The thermionic cathode filament is heated and emits electrons. The electrons are accelerated by a grid into the ionization region. The electrons collide with and ionize ambient neutral particles that enter the gauge. Those ions are collected by a grounded ion collector and the measured current is linearly proportional to the number density of the gas. The positive ion current provides an indirect measurement of the gas pressure.



**Figure 1. Nude Bayard-Alpert hot cathode ionization gauge.**

### 1) Basic Operating Principles

The pressure measurement via the collection current is a function of the number of molecules per unit volume, the ionization cross section for the gas of interest at the appropriate electron energy, the emission current, and the path length of the electrons.<sup>4</sup> Using the ionization cross section,  $\sigma_i$ , length of the ionizing space,  $L$ , and cross-sectional area of the electron beam,  $A_e$ , we can use the molecular density,  $n$ , to determine the number of molecules in the volume.

$$N_V = n L A_e \quad (1)$$

Using the ideal gas law:

$$n = \frac{P}{k T} \quad (2)$$

The total ionization cross sectional area of the molecules within the volume is:

$$A_\sigma = n L A_e \sigma_i = \frac{L A_e \sigma_i P}{k T} \quad (3)$$

and the fraction of incoming electrons that participate in the ionizing collisions is:

$$\frac{A_\sigma}{A_e} = n L \sigma_i = \frac{L \sigma_i P}{k T}. \quad (4)$$

If ‘N’ is the number of electrons entering the anode grid per unit time, then the number of collisions per unit time is:

$$\frac{N L \sigma_i P}{k T}. \quad (5)$$

Finally, assuming efficient collection and electron charge ‘e’, the collector current will be:

$$I_c = N L \sigma_i \left[ \frac{P}{k T} \right] e. \quad (6)$$

Substituting for the electron emission current,  $I_e = N e$ , we obtain the expression:

$$I_c = \left[ \frac{L \sigma_i}{k T} \right] \cdot e \cdot P. \quad (7)$$

The first term of equation 7 is a function of the gas type, the geometry of the gauge and the temperature; typically defined as the gauge sensitivity factor ‘S’; yielding the standard ionization gauge equation:

$$I_c = S I_e P \quad \text{or} \quad \text{Sensitivity} = \frac{\text{Ion Current}}{\text{Electric Current} \cdot \text{Pressure}}. \quad (8)$$

It is the linear relationship between pressure and the ion current that allows the Bayard-Alpert gauge to provide an accurate continuous indicator of pressure of the range of interest to electric propulsion testing. The BA gauge will deviate from linearity at higher pressure, but not below  $10^{-4}$  Torr. The linear range can be extended to higher pressures by lowering the electron emission current. There are several methods and factors for using BA gauges at higher pressures, but are beyond the applicability of this document. It should also be noted that this linearity at low pressures allows the BA gauge to be calibrated in the linear region and extend its performance range below the calibration levels. This methodology has been confirmed down to  $10^{-9}$  Torr.<sup>5</sup>

### 2) Gas Dependency

The BA gauge sensitivity is directly related to the value of the ionization cross section of the corresponding gas molecules and the sensitivity factor of the gauge is only valid for the gas specified, the pressure readout only provides a direct reading for that gas. The standard gas by industry is nitrogen. All gauges include relative sensitivity factors,  $R_g$ , and are commonly referred to as gas correction factors, the typical correction factors are noted Table 2 in Section IV-C. Gas correction factors are pressure dependent and can become unreliable above  $10^{-5}$  Torr. While this is not a large concern for xenon in the range of electric propulsion testing, higher precision is obtained through calibration against the specific gas using during testing.

### 3) Bias Voltage and Emission Current Dependence

Collector current is a function of the electrode potentials because both the electron trajectories and ionization efficiencies depend on these voltages. Table 1 provides recommended gauge potentials. The sensitivity of an ion gauge can change by 0.1%/V and 1%/V for filament-to-grid and filament-to-ground variations in potential, respectively.

**Table 1. Recommended electrode potentials.**

Collector Potential	0 V <sub>dc</sub>
Filament Bias	+30 V <sub>dc</sub>
Anode Grid Bias	+ 150-180 V <sub>dc</sub>
Shield Potential	0 V <sub>dc</sub>

The recommended positive filament bias is sufficient to assure that no electrons contact the ion collector; only 5 V<sub>dc</sub> is required. Increases in collector voltage are undesired and decrease the ion current due to reduced path length and electron penetration into the anode grid space and the reduced electron energy. Cathode bias should be applied to the bottom of the filament.

The filament-to-anode voltage determines the collisional energy of the electrons that traverse the inner volume of the grid cage. The electron energy is simply calculated, in eV, as the difference in bias voltage between the anode grid and the filament. The electron energy for the prototypical ion gauge controller is 150 eV. If the collector current is measured for varying grid potentials, at a fixed pressure (above 10<sup>-7</sup> Torr), filament bias and electron current, the curve showing  $I_c$  versus  $V_g$  follows the expected characteristic shape of gas ionization probability versus electron impact energy. The ionization current rises rapidly with  $V_g$  up to 200 V and varies slowly with grid voltages above this value.<sup>6</sup>

Sensitivity is also dependent on the emission current. Emission currents ranging from 0.1 to 1 mA usually show no degradation. Increasing emission currents to 10 mA can decrease sensitivity by 20% and cause non-linearity above 10<sup>-5</sup> Torr. BA gauges should be operated with 1 mA emission current. Higher current should only be used to increase the speed of outgassing. The filament current supply should be DC. The controller should control bias voltages within a few volts directly at the gauge head and emission currents should be controlled to within a few percent.<sup>7, 8</sup> If the bias voltage is controlled within the gauge controller, the voltage can vary with heating current due to resistive voltage drop across the cable and vary with cable length. The filament bias voltage shall be controlled at the gauge head and not the controller box.

### 4) Temperature Dependency

The hot-cathode ionization gauge must obtain thermal equilibrium internally and with its surrounding before accurate pressure measurements can be made. This process can be accelerated with the use of the degassing feature on most commercial hot-cathode ionization gauge systems.

The hot-cathode ionization gauge shall be operating for more than one (1) hour and should operate for two (2) hours before experimental measurements are taken to ensure thermal equilibrium is reached and outgassing of the surfaces is complete. The gauge shall operate for at least one (1) hour after a degas event prior to making measurements.

Studies of temperature effect on results have shown that there is lower practical accuracy impact than theoretical analyses would predict when accounting for both density and thermal transpiration.<sup>9</sup> The envelope temperature of the BA gauge will typically be much higher than ambient due to the power radiated by the hot filament and absorbed by the envelope's walls. The absorption of energy from the filament by the envelope increases with age as the walls get progressively darker due to contamination. Variations in filament work-function and emissivity due to aging, contamination or chemical reaction with the gas will result in changes in filament and envelope temperature that might require correction for accurate measurements.

Studies have shown the ion current to change by 0.075%/K, approximately half of that expected from the envelope temperature. Sensitivity changes can be mitigated through calibration under the same temperature as those during test. If a large variation is expected, a correction may be required. A temperature correction methodology is provided in the Appendix.

### 5) Magnetic Field Dependency

Magnetic fields near BA gauges can significantly impact performance by affecting the trajectories of the charged particles. The effects are non-linear with both magnetic field and pressure. However, no conclusive study has been completed to characterize the magnetic field effect on pressure measurement accuracy. If following gauge

placement recommendations in section IV, the performance change should be negligible. If a gauge is placed relatively close to a magnetic field source, testing the gauge with and without the expected magnetic field is recommended.

## B. Gauge Construction

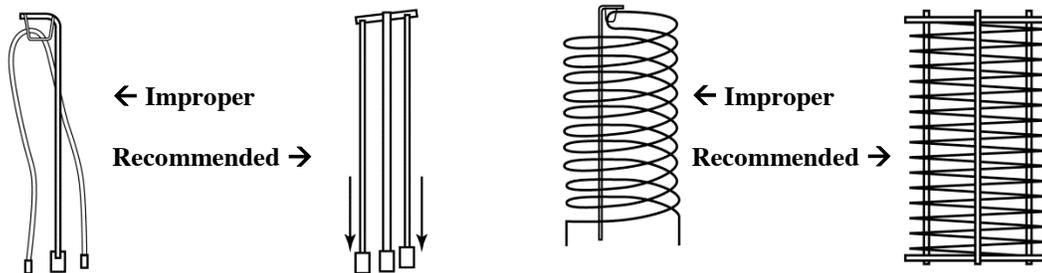
The standard Bayard-Alpert hot-cathode gauge is the most prevalent pressure gauge used during electric propulsion testing. The hot-cathode BA gauge is generally considered the most accurate continuous indicator for pressure below  $10^{-3}$  Torr. Vendors specify typical accuracies of  $\pm 20\%$ , but 30-50% total uncertainty has been historically demonstrated, and the gauges exhibit limited stability required for test-to-test repeatability. A study on long-term stability (580 days) of BA gauges observed changes in calibration that ranged from -57% to 72%.<sup>10</sup> Follow-on studies highlighted the influence of the controller on uncertainty, and with a quality controller and proper calibration procedures, accuracies of  $\pm 20\%$  can be achieved. However, design and construction techniques can significantly improve both the accuracy and stability of the pressure measurements.<sup>11</sup> Individually calibrated high accuracy gauges offer accuracies better than 4% over the range most applicable to electric propulsion testing.

### 1) Gauge Geometry Dependency

The key geometry factors of gauge design include the filament to grid spacing, the collector wire location and diameter, the anode grid end closures and the grid diameter. It is also the variations and change in these parameters over time that impact gauge-to-gauge variability and stability, respectively.

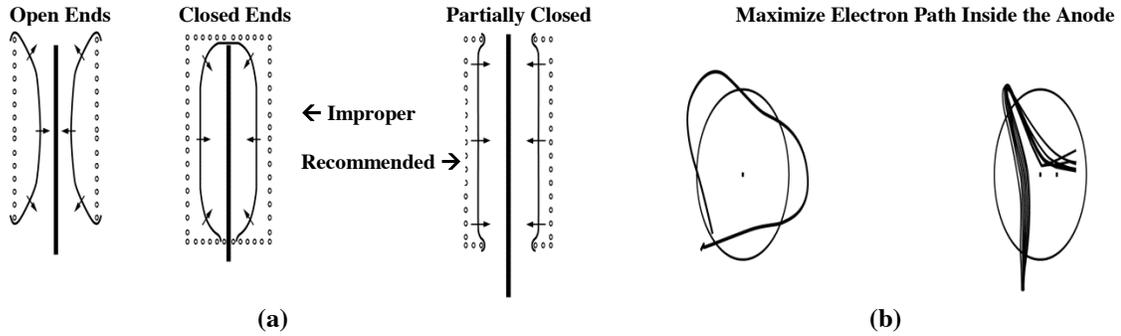
Standard BA gauges are known to exhibit large inaccuracies and instabilities due to gauge construction limitations. BA gauge filaments and grids can sag or change shape due to thermal cycling and even placement. Preferred mounting orientation is with the filament and anode grid in a vertical position to minimize the electrode distortion caused by gravity pull and thermal cycles. Gauges with opposed tungsten filaments have better long-term stability by a factor of two. One should use all-metal gauges if there will be helium leak testing due to the heated glass permeation of helium. Gauges shall be designed to support both the filament and the acceleration grid. Supported elements can eliminate the preference for vertical mounting and reduce gauge placement limitations as shown in Fig. 2. Tight cathode filaments stretched between rigid posts have longer term stability over hairpin shaped or unsupported ribbon cathodes.

Another source of variations is the electrode position precision. Variations in placement can impact the electron collection resulting in measurement inconsistencies. To maintain stability over the entire pressure range, it is also desired to have a thicker diameter ion collector. The thicker wire provides increased mechanical stability and higher overall sensitivity. Increasing the collector wire diameter from 0.25 mm to 1 mm increases sensitivity by a factor of 2. The ion collector should be approximately 1 mm in diameter. Ultra High Vacuum (UHV) gauges typically use smaller diameter collectors and should be avoided. The only disadvantage to the thicker (1 mm) wire is a higher sensitivity to the energetic ions formed by electron-stimulated desorption; avoidable through bake-out and degas.



**Figure 2. Supported and unsupported filaments and anode grids.**

Filaments should be sized and placed for emission of electrons in the radial electric field region to minimize the axial field effects on the electron paths as shown in Fig. 3a. The filaments should also be designed to maximize the electron path to be within the anode grid and shown in Fig. 3b. Lastly, the addition of grid end closures to the gauge will prevent the escape of uncaptured ions from the opened ends of the cylindrical grid and increase sensitivity as in Fig. 3a. The sensitivity advantage declines rapidly above  $10^{-5}$  Torr, but open end grids would only be recommended if testing occurred in pressure ranges up to  $10^{-3}$  Torr.



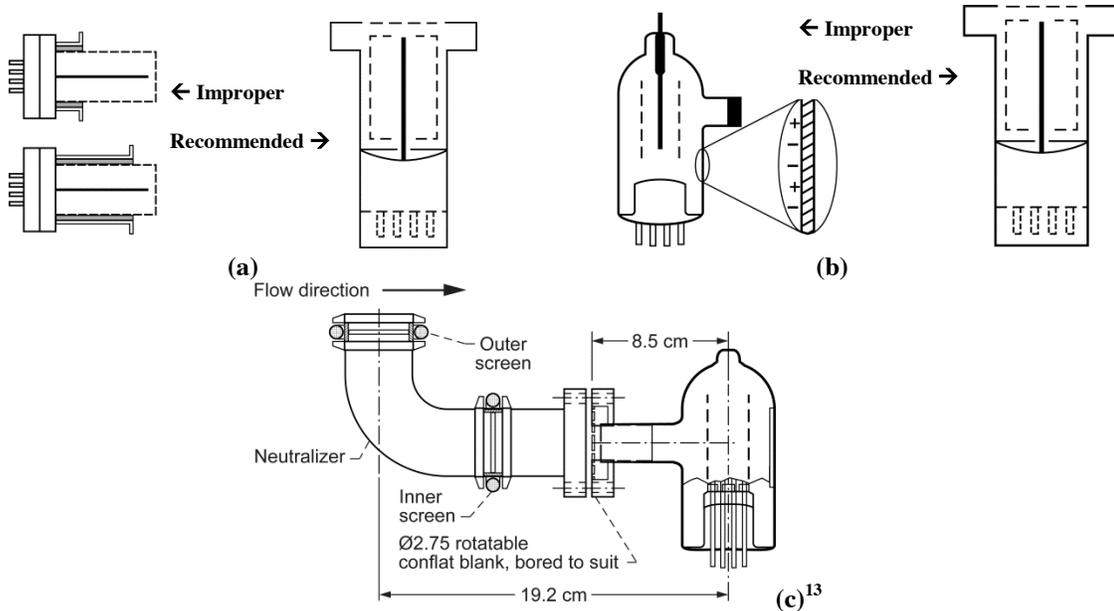
**Figure 3. Filament design recommendations.**

2) *Gauge Envelope Dependence (Nude Gauges)*

The envelope size and shape has been studied extensively for conventional nude BA gauges.<sup>12</sup> The results show that changes in the gauge envelop can result in measurement errors of 50%. Modern high accuracy gauges rely on heavy shielding to protect the electrode from external and uncontrollable fields, better define charged particle trajectories, and improve both the gauge-to-gauge reproducibility and long term stability.

It is critical to maintain a consistent electric environment during testing. Gauges can house the entire electrode assembly in a grounded metal envelope that completely surrounds the structure to provide a stable electric environment for charged article trajectories, Fig. 4a. A grounded perforated high-conductance shield over the port can electrically isolate the electrode structures from the rest of the vacuum system. A grounded conducting shield between the anode and the feed-through also prevent the ceramic insulators from becoming contaminated and charged, Fig. 4b.

The situation may also arise for pressure measurements within the plume of a thruster. The pressure levels will still be in the high vacuum range, thus ionization gauges are necessary. However due to the abundant presence of plume ions, the gauge will have high uncertainty if directly exposed to the plasma. If the gauge must be exposed to the plasma, a neutralizer should be installed on the gauge to prevent discharge ion collection by the gauge. The neutralizer should prevent line-of-sight to the hot plasma and be grounded to provide means for neutralization of discharge ions. Nude gauges can have higher accuracies by eliminating conductance losses, however because shielded gauges can prevent the false measurement of collecting electrons from the plasma generated by the thruster, they may be required. The conductance losses of the neutralizer should be minimized and shall be included in the pressure calculation and gauge calibration. The screen spacing should be sized smaller than 5 Debye lengths so the sheath merges. An example setup is provided in Fig. 4c.



**Figure 4. Recommend gauge envelope designs.**

Commercial products are available that meet the recommendations above such as the Granville-Phillips Stabil-ion Gauge model 370120 with either an IGC100 or Granville-Phillips Series 370 controller. This information is given as convenience and does not constitute an endorsement. Equivalent products may be used if they can be shown to produce consistency results or the uncertainty analysis must reflect the inaccuracies and sources.

### 3) Filament Design

The filament design and material is critical to supply a stable electron emission current source and therefore high accuracy measurements. The desired characteristics of the filaments are to have a reduced chemical reactivity with the rarefied environment to be measured, reduced evaporation rate at the operating temperature for prolonged life, a vapor pressure at least one order of magnitude lower than that measured, and low levels of outgassing.

Filament materials used in BA gauges are either pure metal (*e.g.*, tungsten) or oxide-coated (*e.g.*, thorium-coated iridium). The operating temperature of tungsten cathodes is between 1900 °C and 2200 °C. At this high operating temperatures, contaminating electronegative gases, which would increase the work function and reduce emission levels, are rapidly evaporated from the filament surface. As a result, tungsten filaments provide more stable gauge operation compared to metal oxide cathodes. The lifetime of W filaments, as determined by typical evaporation rates, is typically 1000 hours at  $10^{-6}$  Torr. Tungsten is also not seriously affected by hydrocarbons during operation.

Thorium-coated iridium (ThO<sub>2</sub>Ir) filaments are the oxide-coated cathodes that are the most common filaments manufactured in the US while European filaments usually contain yttrium oxide instead. These cathodes are prepared by depositing a layer of thorium (*i.e.*, thorium oxide) on a base metal of iridium by cathododesis. Iridium is the preferred substrate because it is very resistant to oxidation and does not burn out if exposed to high air pressures while hot. ThO<sub>2</sub>Ir filaments are very resistant to poisoning and BA gauges with ThO<sub>2</sub>Ir filaments are known to survive several brief exposures to atmospheric air, without any performance deterioration.

The preferred material choice is dictated by application. Many of the material advantages are exhibited in the range outside the interest of electric propulsion testing,  $<10^{-8}$  Torr or above  $10^{-4}$  Torr, and both are sufficient for electric propulsion testing.

## III. Calibration

The National Institute of Standards and Technology (NIST) calibration for high vacuum (as low as  $10^{-9}$  Torr), uses a known gas that flows into the top of a vacuum chamber, passes through an orifice in the middle, and exits at the bottom. Kinetic theory allows the conductance of the orifice to be calculated from its known diameter, which in turn allows the pressure drop to be calculated accurately from the conductance and flow rate. Calibration by a gauge manufacturer is typically performed using a NIST reference spinning rotor gauge, a similar technique can be used during recalibration.

Calibration of the gauge shall be performed with NIST traceability with resulting uncertainties carried through the pressure measurement data reduction. The calibration should occur down to pressures at least one order of magnitude lower than expected during the range of testing.

With proper calibration, vendors specify accuracy with air better than  $\pm 4\%$  and repeatability  $\pm 3\%$ . Testing has indicated an uncalibrated accuracy of 6% and if calibrated on xenon, better than 3% accuracy. However, using the recommended gauge design and practices alone is still insufficient for accuracies better than  $\pm 20\%$  without a quality controller. Experience using the Stabil-ion gauge without the controller also calibrated to NIST standards has shown comparable accuracies to standard BA gauges. The cabling, feed-through lengths, and connector quality can also impact measurement accuracy.

Gauges should be calibrated to a NIST standard with the controller electronics, feed-throughs and cabling, and gauge in the configuration to be used. Calibration should be performed on the primary gas, *e.g.*, xenon, if practical.

In general, the reproducibility of the hot-cathode gauges can be as good as 2% over a year of continuous operation in controlled vacuum conditions. A study of successive calibrations of 20 tube hot-cathode gauges with tungsten filaments showed the standard deviation of the maximum difference between successive calibrations was 3%, with a maximum of 12%.<sup>14</sup> The sensitivity of the gauges tended to decrease in most cases. The long term stability of the pressure measurement is strongly affected by its operational history and gas exposure. Corrosive gases can quickly degrade the accuracy of BA gauges. Calibration intervals should be the lessor of twelve (12) months or 2,000 hours of non-continuous operation.

## IV. Setup and Measurement

Initial pumping of the facility should occur without operating the ionization gauge. The gauge may be turned on after the facility reaches  $5 \times 10^{-4}$  Torr. The ionization gauge is typically mounted at the vacuum chamber wall. The

assumption is the distribution of particles everywhere in the chamber can be characterized and thus measurement at one location is sufficient.

#### **A. Pressure Measurement and Gauge Location**

The placement of the vacuum gauges greatly affects their uncertainty in measuring the background pressure. The uncertainty in vacuum pressure measurements arise from two sources: uncertainty inherent in the gauge, and varying pressure throughout the vacuum system. The uncertainty inherent in the vacuum gauge will usually be stated by the manufacturer and is unavoidable. The operating history of a hot-cathode gauge and the ambient environment in the vacuum system causes the gauge to act as either a source (outgassing) or sink (surface pumping) of gas. Typical 1-mA emission filaments require 10-15 W of input power. This amount of power is sufficient to cause thermal degassing from the gauge elements and surroundings. The significance of these effects depends of the characteristics of the vacuum system, *i.e.*, a small ultra-high vacuum system with low pumping speed is very sensitivity to external sources and sinks of gas.

The second source of uncertainty comes from the pressure distribution in the chamber caused by localized placement of vacuum pumps and pumping surfaces. The pressure will be lower near the pumps. With an operating thruster, the pressure will also be high in the immediate vicinity of the discharge plume. Recommendations and requirements are provided to provide the highest pressure correlation to a relevant reference location with minimal uncertainty.

The pressure gauge shall be mounted inside the vacuum chamber and have line-of-site with the thruster. The pressure gauge should be mounted near the chamber wall at the exit plane of the thruster, located at least 0.6 chamber radii away from the centerline and at least one (1) meter from the outer diameter of the thruster. The pressure gauge can be standard flush mounted to the port, but should not be mounted in a recessed cavity.

Limitations can make it impossible or inconvenient to mount the pressure gauge(s) along the wall at the exit plane. However, the pressure measurement shall be reported as the pressure near the chamber wall at the exit plane of the thruster and sufficient facility description and gauge location shall be provided to allow the calculation of the pressure near the wall at the exit plane of the thruster. Additional uncertainties must be included due to the facility configuration, complexity of the geometries, distance from the exit plane, and method of calculating the pressure from the measurement location to the reference location.

If the pressure gauge cannot be located near the wall at the exit plane of the thruster, it should be located near the wall nearest the exit plane as practical, in front of the thruster, and before exposure to a 60° half angle of the thruster channel wall nearest the gauge. The number of pumping surfaces between the exit plane of the thruster and the gauge should be minimized.

If the gauge(s) cannot be located downstream prior to exposure to the 60° half angle, the gauge(s) shall be at least two (2) meters downstream of the thruster exit plane. Additional shielding may be required as discussed previously when the gauge will be exposed to the plasma. The additional shielding and impact on measurements shall be characterized, *e.g.*, through a combination of both hot-fire and cold flow testing.

Pressure measurement shall be made after facility achieves steady-state and no less than two (2) minutes after a commanded flow-rate change greater than 10%. Commercial controllers have a typical sampling rate of 60 Hz. The sampling rate should be no less than 10 Hz. To filter inherent noise, pressure should be measured as the three (3) second average value.

It is recognized that pumping surfaces are often asymmetric near the thruster location due to facility design and limitations. Ideally radial pumping asymmetry should occur at least 2-m downstream of the thruster; as far as practical. However, due to conductance limitations, pumping surfaces are desired near the thruster exit plane.

Radial asymmetric pressure differentials should be minimized. Asymmetric pressure variation shall be less than 20% at the thruster exit plan one (1) thruster radii from the out exit channel wall. Performance, plume measurements, and life testing impacts of radial pressure asymmetries >10% shall be characterized based on specific test objectives.

#### **B. Facility Effect Characterization**

Due to limited data and relative low cost of augmenting electric propulsion testing, it is recommended that all future electric propulsion testing include background pressure effects characterization when practical. Baseline testing should occur at the maximum capability of the facility, and then measurements with the same diagnostics and methodology should be repeated with artificially increased background pressure. The background pressure range should encompass the background pressure expected over the life cycle of the thruster. Background pressure should not be increased through mechanical pumping reduction of an on-going test. Background pressure should be increased through supplemental supply of the primary gas (*e.g.*, xenon) downstream of the thruster. There should be

at least one pumping surface between the additional gas supply and the pressure gauge. The supplemental gas flow should be at least two meters downstream of the thruster exit plane or preferably near the centerline of the facility.

There is no absolute requirement on the position of the injected gas provided the environment is sufficiently characterized to discern pressure distribution and minimize differential pressure zones near the thruster and pressure gauge(s).

### C. Calculations and Other Measurements

#### 1) Reference Pressure

If the measurement cannot be made directly, a calculation of the reference pressure is required. The reference pressure calculation can range from simple conductance and Clausing transmission corrections to high fidelity facility pressure distribution simulations.<sup>15</sup> All assumptions necessary to independently reproduce the calculations shall be recorded. Example calculations are provided in the appendix.

The relevant reference pressure shall be near the wall at the exit plane of the thruster. The pressure shall be corrected for the primary gas (*e.g.*, xenon correction factor of 0.348). While calibration and measurement using the gas of interest is preferred, a linear correction factor over the pressure range of interest is generally sufficient. Table 2 is a list of applicable correction factors.

**Table 2. Common gas correction factors.**

Xe	0.348
Kr	0.515
Ar	0.775
H <sub>2</sub>	2.17

#### 2) Neutral Density

Average neutral density can be obtained using the ideal gas law where  $P$  is the pressure,  $n$  is the number density,  $k$  is Boltzmann's constant, and  $T$  is an average temperature. The temperature should be taken as the temperature of the chamber wall at the location of the pressure gauge.

#### 3) Effective Pumping Speed

Facility pumping speeds are often reported based on the total additive capability of mechanical and cryogenic pumping systems. This method greatly exaggerates a facilities pumping capability, does not account for conductance losses, Clausing transmission limitations, or basic geometry/configuration impacts to pumping performance and contributes to facility-to-facility comparison inconsistencies.

Facility pumping speed shall be reported as the effective pumping speed at the exit plane of the thruster. Effective pumping speed should be calculated directly from the reference pressure calculation/measurement. This serves as a measurement of effective pumping speed.

$$P_{Effective} = \frac{\dot{m}}{p_{measured} - p_{base}} \quad (9)$$

The expected effective pumping speed can also be calculated using the vendor specified performance, but must include all performance losses from the pump to the facility (*i.e.*, pump extension/nozzle, baffle, chevron, etc.) and facility losses. Rated pumping speeds also vary as a function of pressure and should be considered in facility calculations. Example calculations are provided in the appendix.

## V. Pressure Requirements

Facility effects impact various measurements differently. Performance is primarily affected by ingested propellant background gas; which can be difficult to estimate due to "beam pumping" effects. Lifetime is primarily affected by the changes in the ion fluxes that cause erosion of the various thruster surfaces. Far-field plume measurements are most sensitive by plasma densities and energies caused by the facility environment. Early testing of the Stationary Plasma Thruster (SPT)-100 provided guidance regarding acceptable facility pressure requirements for performance, lifetime and near-field plume measurements;  $5 \times 10^{-5}$  Torr,  $5 \times 10^{-6}$  Torr, and  $1.3 \times 10^{-5}$  Torr, respectively.<sup>16, 17</sup> The authors note that the basis for this accepted standard for HETs does not include any specifications regarding pressure measurement methodology, locations, correction factors or even the applicability to thrusters beyond the SPT-100.

Extremely low levels of contaminants can have profound effects on measured lifetimes. Nitrogen and air can interact with surfaces and drastically change their sputter rates. Also, there is limited data on background partial pressures. For gridded ion engines, it has been observed that breakdown rates of ion thruster begins to be effected in the low  $10^{-5}$  Torr range and lifetime measurements may be impacted at  $5 \times 10^{-7}$  Torr.<sup>18</sup> However, with the NSTAR life test conducted at  $4 \times 10^{-6}$  Torr and nominal flight mission performance; it is likely to become the accepted

standard for life testing of gridded-ion engines.<sup>19</sup> The NEXT Long Duration Test has been conducted at a maximum pressure of  $2.5 \times 10^{-6}$  Torr<sup>20</sup> and the BPT-4000 Life Test was conducted at  $2.5 \times 10^{-5}$  Torr.<sup>21</sup>

Several past studies have highlighted thruster performance and various diagnostic measurement sensitivities to facility background pressure. There is insufficient test data for absolute requirements for all thruster designs, scales, and measurements of interest. This section provides recommendations as an initial position, but is acknowledged to be incomplete and subject to change as additional testing and analyses warrants. Pressure requirements below are at the reference location and are corrected for the primary gas.

#### A. Hall Thrusters

Performance for Hall thrusters shall be measured with background pressure  $\leq 3 \times 10^{-5}$  Torr. Near-field plume measurements shall be made with background pressure  $\leq 1.3 \times 10^{-5}$  Torr. Life testing of Hall thrusters shall be conducted with a background pressure  $\leq 5 \times 10^{-6}$  Torr.

#### B. Gridded-Ion Thrusters

Performance for gridded-ion thrusters shall be measured with background pressure  $\leq 3 \times 10^{-5}$  Torr. Near-field plume measurements shall be made with background pressure  $\leq 1.3 \times 10^{-5}$  Torr. Life testing of gridded-ion thrusters shall be conducted with a background pressure  $\leq 5 \times 10^{-6}$  Torr. Additionally, testing shall validate contaminant partial pressures  $\leq 5 \times 10^{-7}$  Torr do not impact erosion flux rates.

### VI. Cautionary Notes

#### A. Correlation to Reference

The most critical concern with pressure measurements and effective pumping speeds is the proper correlation to the reference plane and sources of discrepancies. Consider a 60 foot long facility with a 15 foot diameter with the far end cap open to space (*i.e.*, infinite pumping capability). Regardless of pumping capability, the facility has a maximum conductance of 889,000 l/s. Placing the thruster 1 meter from the near-wall and pressure gauges at two different locations highlights the importance of conductance and the need to correlate to the reference plane.

Testing a moderate 4.5-kW thruster in this configuration, with infinite pumping speed, will not meet the required background pressure for performance testing. Also, the two wall gauges, if used directly would indicate the facility does have sufficient performance; shown in Fig. 5. Most importantly, either gauge, if correlated correctly could be used to calculate the reference pressure and effective pumping speed. The emphasis is not on limitations of where measurements are performed, rather that they are correlated to an appropriate and relevant reference.

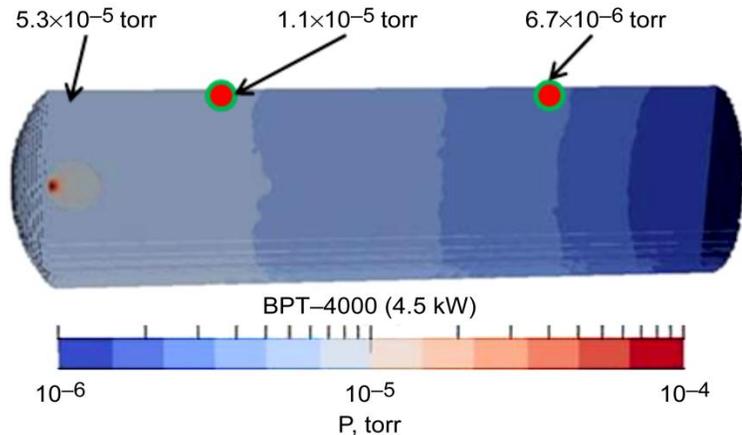


Figure 5. Hypothetical test facility with infinite pumping capability.

#### B. Gauge History Dependence

The dependence of the sensitivity drifts on the type of gauge and its operating conditions has made it impossible to develop a unified model or theory that completely and systematically explains all experimental observations. Most knowledge is phenomenological and based on the experience accumulated over several decades of pressure measurements with commercial BA gauges.

Instabilities in commercial ionization gauges can often be traced back to changes in the path of the electron beam caused by several different aging effects.<sup>22</sup> Most ion gauge controllers do an adequate job at maintaining the electron emission current and bias voltages at a constant value; however, they have no influence over the trajectories of the electrons once they leave the hot filament surface.

Changes in the emission characteristics of the filament are of high concern since they directly affect the electron trajectories and can result in changes in both the potential distribution and the charged particle trajectories inside the anode grid.<sup>23</sup> Large variations in the emission characteristics of the filament can be caused by the following effects:

1) *Changes in geometry of the electrode structure*

This effect is often due to either repeated thermal cycling and/or mechanical shock and can be mitigated through the use of tensioned filaments and electrode supports.

2) *Local temperature variations in the filament wire and changes in the cathode dimensions*

Filaments will degrade over time and exhibit continuous metal evaporation during emission; thinning the filament. The filament is usually the life limiting element of a BA gauge. As the filament becomes thinner, the controller will increase the filament temperature to maintain current with the reduced surface area.<sup>24</sup> Preferential depletion of the central portion of the filament will cause changes in the filament shape and temperature distribution; resulting in a change in electron emission distribution along the filament. The filament should be visually inspected during each calibration cycle.

3) *Surface contamination and chemical reaction*

4) *Changes in the filament coating*

Many cathodes have deposits of low work-function layers on refractory metal wires with potential to degrade or detach during use. Evaporation and ion bombardment may also deplete the central portion of the coating, resulting in differential emission shifting towards the ends of the filament.

### **C. Leakage Currents**

Hot-cathode ionization gauges operate through the collection of small currents, and therefore leakage currents can significantly impact performance and should be mitigated. Leakage current may occur around the electrical connection to the gauge pins. Plasma electrons can enter in the gap between the plug and gauge and effect measurements. To prevent this leakage current, the connection point should be covered in an insulating tape. The area around the collector pin on the gauge must also be kept clean at all times on both the air and vacuum sides of the feed-through connectors. It is also important to have quality leads to make good connections to the controller.

### **D. Outgassing**

The outgassing of hot cathode gauges is a potentially large source of error when such gauges are used at base pressure levels in high vacuum systems. Outgassing levels are particularly high when a gauge is turned on for the first time after exposure to ambient or high gas pressures. Degassing and/or bake-out can reduce the outgassing during testing. The recommended method to reduce gauge outgassing is to bake out the gauge for an extended period of time. BA gauges are typically degassed after the gauge has been exposed to ambient or if surface contamination is suspected. The BA gauge should not be used for at least one (1) hour after degassing, but multiple hours are preferred. The NIST High Vacuum Group, concerned mostly with pressure ranges near electric propulsion base pressures and below, recommends the elimination of the degassing practice. Observations have shown that gauges effectively degas during normal operation while the vacuum system is baked. Their recommendation is to only degas if the gauge is heavily contaminated or after exposure to active gases such as oxygen and to degas at the minimum current for a longer period of time.<sup>25</sup> Modern controllers allow for degas power and time variability.

## **VII. Summary**

Facility background pressure is known to impact the performance, Hall thruster acceleration zone, near-field plume measurements, etc. Various thrusters and designs have also shown inconsistent trends in diagnostic measurements due to background pressure. Thrusters will be tested at various facilities and conditions throughout research, development, qualification, system integration, and mission operations. The community currently lacks sufficient test data and analyses to analytically correct for all facility effects. The literature archives include significant test efforts without pressure measurements provided at a relevant reference or sufficient information to calculate independently. Existing practices rely on gauges known to have inaccuracies several factors higher than achievable with proper hot-cathode gauge construction, NIST controllers and calibration. Without community consensus on pressure requirements, it is difficult to advocate for facility investments, or worse, inconsistencies may question the validity of key results from costly tests. This document is only a preliminary step towards pressure measurement standardization and it is greatly anticipated the community will continue additional testing to help refine and define additional pressure requirements and facility effect characterization.

## Appendix – Example Calculations

### A. Effective Pumping Speed via Reference Pressure – Recommended Method

Assume the facility base pressure was measured as  $1 \times 10^{-8}$  Torr prior to thruster operation. A xenon thruster is being tested with a total (anode plus cathode) flow rate of 150 sccm. The pressure near the wall at the exit plane of the thruster is measured as  $1.3 \times 10^{-5}$  Torr. Determine the effective pumping speed of the facility at this set point.

First convert the flow rate from sccm to Torr l/s.

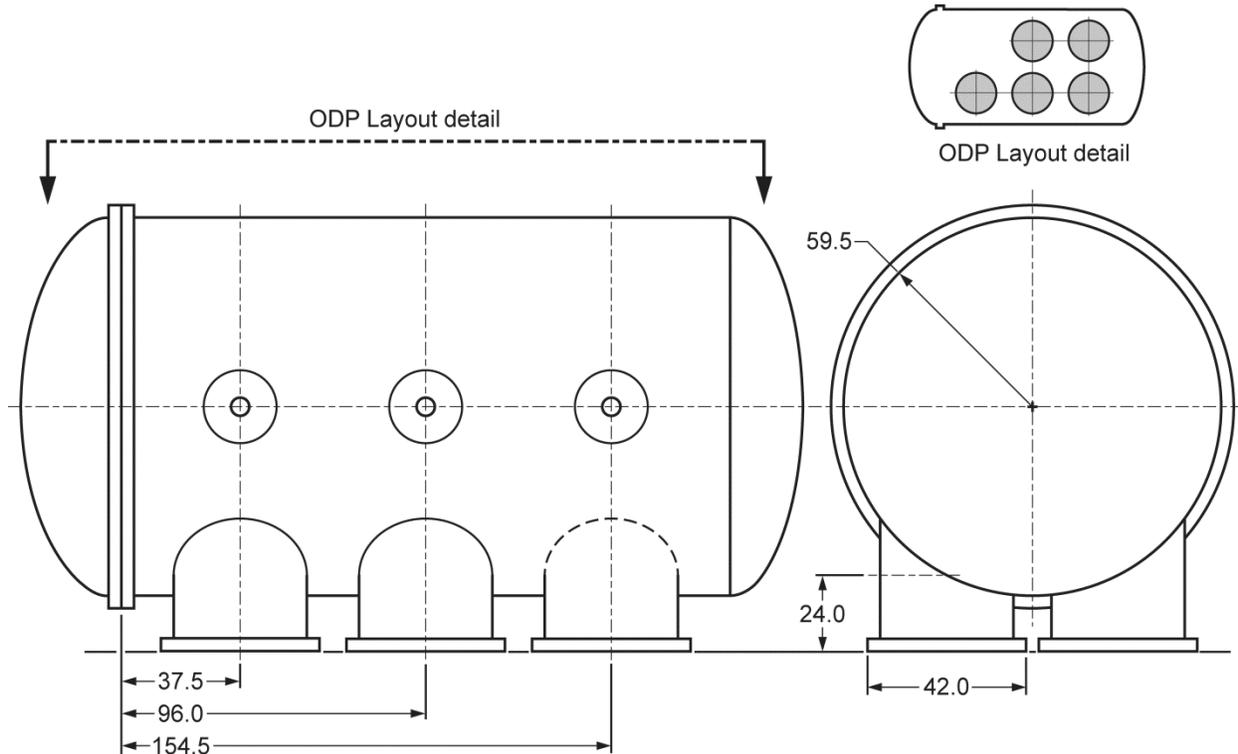
$$150 \text{ sccm} = \frac{760 \text{ torr} \cdot 10^{-3} \text{ l}}{60 \text{ s}} = 1.9 \frac{\text{Torr l}}{\text{s}} \quad (\text{A1})$$

Then applying equation 9:

$$P_{\text{Effective}} = \frac{1.9 \frac{\text{Torr l}}{\text{s}}}{1.3 \times 10^{-5} \text{ Torr} - 1.0 \times 10^{-8} \text{ Torr}} = 146,266 \frac{\text{l}}{\text{s}} \quad (\text{A2})$$

### B. Effective Pumping Speed via Facility Description – Oil Diffusion Pump Facility

To illustrate the procedure for calculating the effective pumping speed at any location in a facility through description, NASA GRC Vacuum Facility-7 is used. NASA GRC's VF-7, is a 10' Diameter x 15' Length Facility with a historical base pressure of  $1 \times 10^{-7}$  Torr and rated with a nominal pumping speed of 125,000 l/s (air) using five 32" Oil Diffusion Pumps (ODPs). The facility schematic and pump locations are illustrated in Fig. B-1. For calculations, the radius of the facility is 59.5 inches for an area of 11,122 in<sup>2</sup> (71,754.8 cm<sup>2</sup>). The length of the facility is 118 in. with ODP 1, ODPs 2 and 3, and ODPs 4 and 5 located at 37.5 in., 96 in., and 154.5 in., respectively.



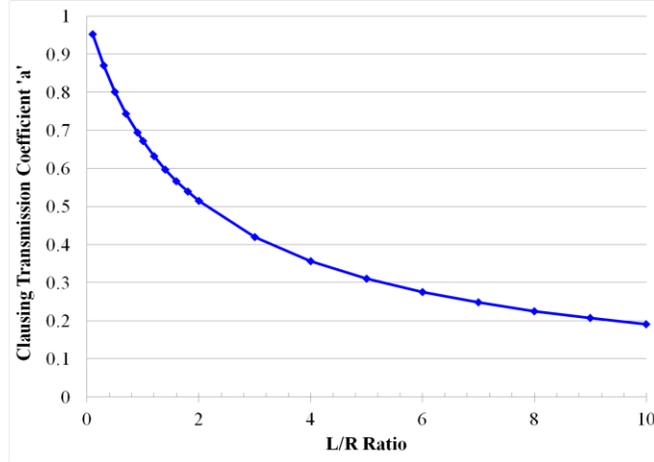
**Figure B-1. NASA GRC VF-7 layout with dimensions in inches.**

Most facilities of interest have a cylindrical shell design. Calculations for conductance in a pipe are straightforward and often used. However, the pipe equation yields its best accuracy when the length-to-diameter (L/D) ratio is greater than 10 while most facilities have L/D ratios less than 5. To accommodate this lower ratio, it is

recommended to calculate aperture conductance and then applying the Clausing Transmission Coefficient for the length-to-diameter component. Molecular conductance is provided in Eq. (B1).

$$C_M = I_R * a * A \quad (B1)$$

Clausing Transmission Coefficients are dependent on the length-to-radius ratio, as shown in Fig. B-2.



**Figure B-2. Clausing transmission coefficients.**

The flow rate across the aperture is dependent on gas species since molecular velocities are dependent on molecular weight. A molecular impact rate is established for specific gas species using the equation below:

$$I_R = P * 3.5 \times 10^{22} / \text{sqrt}(M*T) \quad (B2)$$

Calculations included here are for air and xenon since they are the most relevant to provided pump data and standard Hall and gridded-ion operation, respectively. The base pressure used is dependent on the base pressure of the particular chamber recorded during testing. A typical ODP installation will have a circular extension / nozzle extending from the main vessel terminating with a flange for bolting. The conductance for this nozzle will determine the delivered pumping capability to the main chamber. As required by ISO Standard 1608, the gauge tube is mounted at D/2 above the connecting flange of the pump, and for AVS 4.1, the mounting distance is D/4. The gauge location dimension will be subtracted from the overall length of the nozzle when calculating the conductance.

The nozzle extension also houses a chevron baffle directly above the ODP, or in some cases the chevron is bolted directly to the ODP. If the conductance of the chevron is provided by the manufacturer, then the pumping speed at the chevron outlet can be found using Eq. (B3).

$$1/P_{\text{Eff}} = 1/P_S + 1/C_{\text{Baffle}} \quad (B3)$$

Because the baffle is not mounted at the gauge distances listed in either ISO 1608 or AVS 4.1, an equivalent pump speed at the opening of the pump is necessary for the calculation. Again, we use the Clausing Transmission Coefficient to find the equivalent pump speed. For D/2 (R) and D/4 (R/2), the coefficients are 0.671984 and 0.801271, respectively. For VF-7, the ISO pumping speeds are rated for air at 23,000 l/s mounted according to ISO 1608, therefore the equivalent speed at the pump entrance is given by Eq. (B4).

$$P_{\text{Eq}} = P_S / a = (23,000 \text{ l/s}) / 0.671984 = 34,227 \text{ l/s} \quad (B4)$$

It is also common practice for the baffle manufacturers to provide a pumping speed reduction factor, k. For VF-7, the chevron manufacturer provided a correction factor of 50%. This can be used to determine the conductance of the baffle.

$$P_{\text{Eff}} = k * P_{\text{Eq}} \quad (B5)$$

$$1/(k * P_{Eq}) = 1/P_{Eq} + 1/C_{Baffle} \quad (B6)$$

Therefore:

$$C_{Baffle} = k * P_{Eq} / (1 - k) \quad (B7)$$

For the chevron,

$$C_{Chevron} = 0.5 * 34,227 \text{ l/s} / (1 - 0.5) = 34,227 \text{ l/s (air)} \quad (B8)$$

The halo baffle manufacturer provides a correction factor of 60%.

For the halo baffle,

$$C_{Baffle} = 0.6 * 34,227 \text{ l/s} / (1 - 0.6) = 51,341 \text{ l/s (air)} \quad (B9)$$

The example chamber has a 24 in. nozzle length and a 42 in. nozzle diameter; yielding a Clausing coefficient of 0.643587. The conductance of the nozzle is given by Eq. (B10).

$$C_{Nozzle} = V * A * a \quad (B10)$$

The volumetric flow rate can be found using the perfect gas equation and surface impact calculations provided in Eqs. (11) and (12), respectively.

$$PV = nRT \quad (B11)$$

$$I_S = 3.5 \times 10^{22} * P / (M * T)^{1/2} \quad (B12)$$

Assuming a temperature of 293 K, M for air and xenon as 28.96 and 131.29, respectively, and the measured base pressure of  $3 \times 10^{-7}$  Torr:

$$I_S (\text{Air}) = 3.5 \times 10^{22} * 3 \times 10^{-7} / (28.96 * 293)^{1/2} = 1.1399 \times 10^{14} \text{ atoms} / (\text{cm}^2 \text{ s}) \quad (B13)$$

$$I_S (\text{Xe}) = 3.5 \times 10^{22} * 3 \times 10^{-7} / (131.29 * 293)^{1/2} = 5.3535 \times 10^{13} \text{ atoms} / (\text{cm}^2 \text{ s}) \quad (B14)$$

Then using Eq. (15);

$$V = (I_S / 6.022 \times 10^{23}) * R * T / P \quad (B15)$$

Yielding:

$$V = 11.529 \text{ l} / \text{cm}^2 \text{ s (Air)} \text{ and } 5.415 \text{ l} / \text{cm}^2 \text{ s (Xe)} \quad (B16)$$

Finally, substituting into Eq. (10) for Xe:

$$C_{Nozzle} = 11.529 * 8938 * 0.643587 = 66,322 \text{ l/s (Air)} \quad (B17)$$

The system conductance for the single ODP at the vessel entrance is:

$$1/C_{Total} = 1/C_{Chevron} + 1/C_{Baffle} + 1/C_{Nozzle} \quad (B18)$$

Therefore:

$$C_{Total} = 1/(1/51,341 + 1/34,227 + 1/66,322) = 15,681 \text{ l/s (Air)} \quad (B19)$$

Using the effective pumping speed and nozzle conductance, the pumping speed at the entrance to the chamber can be calculated as:

$$1/P_{Entrance} = 1/P_{Eff} + 1/C_{Total} \quad (B20)$$

Therefore;

$$P_{Entrance} = 10,754 \text{ l/s (Air)} \text{ and } 5,058 \text{ l/s (Xe)} \quad (B21)$$

From these calculations, we can predict what gauges should read from various locations in the chamber using superposition of each of the pumps individual speeds with respect to the conductance calculated for the distance to the pump. Assume the thruster is located 50 cm (~20 in) from the end cap and thrusting into the chamber at a location before the plane of the first pumping surface, as is often the case. The distances from pump 1, pumps 2 and 3, and pumps 4 and 5 in combination with the radius of the facility provide the Clausing coefficients of 0.87, 0.62, and 0.48, respectively. Equation 1 is then applied to calculate the conductance and then finally effective pumping from each of the pumps and summed for the total effective pumping speed as shown in Eqs. (B22)–(B25).

$$P_{\text{Eff}} (\text{Pump 1}) = 1 / (1/C + 1/P_{\text{Entrance}}) = 1 / (1/338,121 + 1/5,058) = 4,984 \text{ l/s} \quad (\text{B22})$$

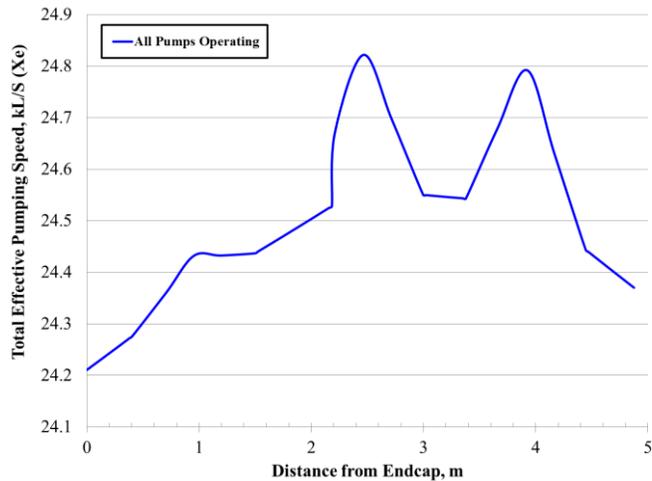
$$P_{\text{Eff}} (\text{Pump 2}) = 1 / (1/239,859 + 1/10,117) = 9,707 \text{ l/s} \quad (\text{B23})$$

$$P_{\text{Eff}} (\text{Pump 3}) = 1 / (1/188,338 + 1/10,117) = 9,601 \text{ l/s} \quad (\text{B24})$$

Then by superposition:

$$P_{\text{Total}} = 24,292 \text{ l/s (Xe)} \quad (\text{B25})$$

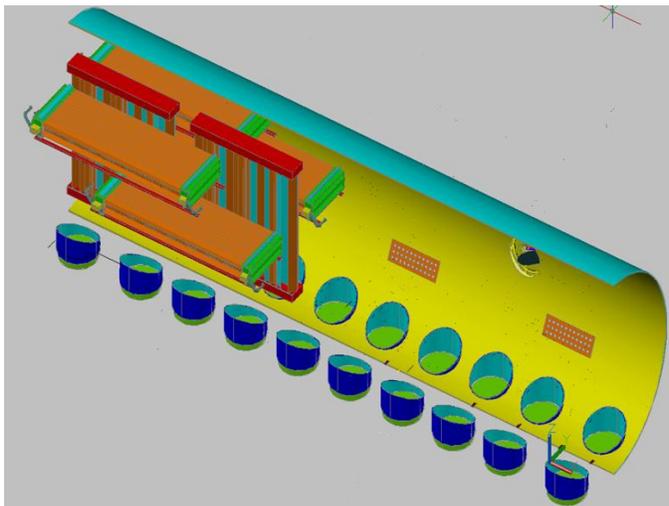
This methodology can be applied to calculate the effective pumping speed from anywhere in the chamber. A plot of effective pumping speed throughout the facility is shown in Fig. B-3. Note that the effective pumping speed is significantly different than the basic sum of the individual pumps. Also, the highest effective pumping speed is achieved by placing the thruster with the exit plane directly over the middle two ODPs.



**Figure B-3. Effective pumping speeds calculated as a function of the location.**

### C. Effective Pumping Speed via Facility Description – with Cryogenic Pumping

Cryogenic pumps can be added as either an external pumping port or as an internal paneled array. For pumps mounted to side port of the main chamber the procedure for assessing the nozzle inlet conductance and effective pump speed is the same as for the oil diffusion pump. In the case of the internal pumping panels, the cross-sectional area of the panel is an obstruction to the molecular flow and must be subtracted from chamber cross-section for an equivalent diameter. An example is the NASA GRC VF-5 with 10 pairs of oil diffusion pumps as shown in Fig. C-1. Using the same methodology as before for ODPs, the effective pumping speed can be calculated for over the length of the facility. Starting farthest

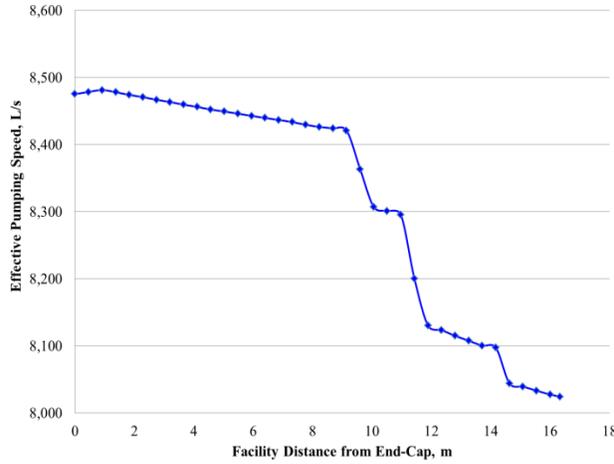


**Figure C-1. NASA GRC VF-5 configuration.**

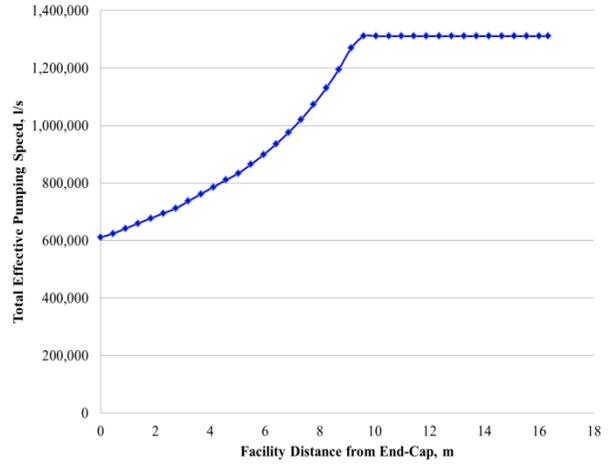
from the cryogenic panels, the effective pumping speed of the first pair of ODPs is shown in Fig. C-2. The figure illustrates the effective pumping speeds peak directly over the pumps and decreasing significantly downstream where the cryogenic panels limit the pumping conductance cross-sectional area.

The two vertical cryogenic panels are 280cm by 300cm and there are four horizontal panels 140cm by 457cm. The panels can pump from front and back surfaces. Using their average pumping speed value, it would indicate a total pumping speed of 4,840,000 l/s. However this cannot be achieved since the total conductance of the 457cm equivalent diameter chamber is only 1,890,200 l/s. The horizontal panels overlap reducing their efficiency to 80%. Using the effective pumping speed calculation, the maximum pumping speed is

calculated to be 1,310,000 l/s (air). Therefore, the ODPs located under the cryogenic panels cannot contribute to the pumping speed of the facility. The effective pumping speed of the ODPs goes to zero and has no pumping influence. This results in a constant pumping speed in the cryogenic pumping region of 1,310,000 l/s (air). The effective pumping speed of the entire facility is shown in Fig. C-3.



**Figure C-2.  $P_{\text{Effective}}$  of ODP Pairs 1 and 2.**



**Figure C-3. Total facility effective pumping speed.**

#### D. Temperature Correction Methodology

Under typical room temperature operating conditions, temperature effects are negligible. However, temperature effects cannot be ignored when gauges are used at or attached to chambers at a wide range of temperatures.<sup>26</sup> The gauge ion current is directly proportional to the neutral density within the gauge.

$$n_{\text{gauge}}\sqrt{T_{\text{gauge}}} = n_{\text{chamber}}\sqrt{T_{\text{chamber}}} \quad (\text{D1})$$

The gauge sensitivity factor during calibration is determined:

$$S_{n_{\text{gauge}}} = \frac{I_c}{n_{\text{gauge}}} = \frac{I_c}{n_{\text{chamber}}\sqrt{T_{\text{chamber}}}} \sqrt{T_{\text{gauge}}} \quad (\text{D2})$$

If the temperature of the gauge is not measured and a gauge sensitivity constant is used; it is related to the gauge sensitivity by:

$$S_{n_{\text{chamber}}} = S_{n_{\text{gauge}}}\sqrt{\frac{T_{\text{chamber}}}{T_{\text{gauge}}}} \quad (\text{D3})$$

If the gauge temperature at calibration is unknown, then the chamber sensitivity constant is not a true constant and it is not possible to determine the gauge sensitivity factor. As shown earlier, if the gas temperature is in a range where condensation does not occur, the gauge sensitivity is independent of temperature and current is only dependent on the neutral density. However, the neutral density in the gauge will depend on temperature of the gauge and the chamber.

$$n_{\text{chamber}} = \frac{I}{S_{n_{\text{gauge}}}\sqrt{T_{\text{chamber}}}} \sqrt{T_{\text{gauge}}} \quad (\text{D4})$$

If a gauge is calibrated in terms of pressure in the gauge envelope,

$$P_{\text{gauge}} = n_{\text{gauge}} k T_{\text{gauge}} \quad (\text{D5})$$

Using the relationship that  $S_{P_{\text{gauge}}} = \frac{I_c}{P_{\text{gauge}}}$  then:

$$S_{P_{\text{gauge}}} = \frac{I_c}{n_{\text{gauge}} k T_{\text{gauge}}} = \frac{S_{n_{\text{gauge}}}}{k T_{\text{gauge}}} \quad (\text{D6})$$

If a gauge is calibrated using pressure in the calibration chamber with a known calibration temperature,

$$I_c = S_{P_{chamber}} P_{chamber} = S_{P_{chamber}} n_{chamber} k T_{chamber} \quad (D7)$$

and

$$S_{P_{chamber}} = \frac{S_{n_{gauge}}}{k T_{chamber}} \frac{n_{gauge}}{n_{chamber}} = \frac{S_{n_{gauge}}}{k T_{chamber}} \sqrt{\frac{T_{chamber}}{T_{gauge}}} \quad (D8)$$

Similarly

$$S_{P_{chamber}} = S_{P_{gauge}} \sqrt{\frac{T_{gauge}}{T_{chamber}}} \quad (D9)$$

Using these relationships, it is now possible to determine temperature correction factors for gauge pressure indications. Correction must be made to the gauge “indicated” pressure shown by the controller. Therefore, the true chamber pressure, with measured temperatures of the gauge and chamber:

$$P_{truth} = n_{truth} k T_{truth} = n_{gauge\ measured} k \sqrt{T_{gauge\ measured} T_{chamber\ measured}} \quad (D10)$$

Then using our relationships and substitution:

$$P_{truth} = P_{indicated} \sqrt{\frac{T_{gauge\ measured} T_{chamber\ measured}}{T_{gauge\ calibration} T_{chamber\ calibration}}} \quad (D11)$$

*Example 1:*

Consider a gauge at room temperature (300 K), calibrated at room temperature (300 K), and attached to a liquid nitrogen-cooled chamber (77 K).

$$P_{truth} = P_{indicated} \sqrt{\frac{300 \cdot 77}{300 \cdot 300}} = 0.506 \cdot P_{indicated} \quad (D12)$$

*Example 2:*

Consider a gauge calibrated at room temperature (300 K), held at liquid nitrogen temperature (77 K) and attached to a liquid nitrogen-cooled chamber (77 K).

$$P_{truth} = P_{indicated} \sqrt{\frac{77 \cdot 77}{300 \cdot 300}} = 0.257 \cdot P_{indicated} \quad (D13)$$

### Acknowledgments

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- <sup>26</sup> Standard Practice for Ionization Gage Application to Space Simulators," ASTM E296-70, Reapproved 2010.